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IN THE SPECIFICATION:

Please amend the first full paragraph on page 3 as follows:

With reference to FIG. 1 and FIG. 2, a stator assembly 30 including an array of cast stator vane segments 32 (only one vane 33 is shown) is located in the engine 20. A rotor assembly 34 includes an array of rotor blades 35 (only two blades are shown) that extends radially outward across the working medium flow path 28. The rotor assembly 34 is supported by various static bearing assemblies within the engine 20. The stator assembly 30 includes an outer case (not shown) that acts as a pressure vessel to confine the working medium gases to the working medium flow path 28.

Please amend the second full paragraph on page 3 as follows:

With particular reference to FIG. 3, each cast stator vane segment 32 has an outer platform 36, an inner platform 37, and one or more vanes ("airfoils") 33 extending between the platforms 36, 37. The inner platform 37 is typically comprised of feet 38 that are disposed circumferentially with respect to the axis of symmetry A. The cast stator vane segment 32 extends between the outer case and an inner shroud 40. The inner shroud 40 has corresponding projections that extend into the grooves formed by the feet 38. By combining a plurality of stator assemblies 30 circumferentially throughout the compressor, high pressure compressor, and turbine sections 22, 23, 26 of the engine 20, the cast stator vane segment 32 interacts with the working medium gases to direct the flow downstream to the rotor blades 35. The stator vane segment can be made from any suitable alloy, such as a high temperature nickel-based alloy.